

CHAP. 2 AIRFRAME S.P. 123B A.L. 1 SHEET 1 OF 6	SERVICING PROCEDURE T55	BAC F53 & T55 (SA) 5A3A Section 1 2nd Edition
Cabin - AI23 Bullet Exhaust Joint	AFSC 42251	TIME EST
Safety and Servicing Notes are to be complied with throughout the work detailed on this card.	43151	
SPECIAL TOOLS AND EQUIPMENT		
Adapter (26DK/95437). Gauge - 0 to 15 lbf/in ² (4F/2191). Gauge - 0 to 10 lbf/in ² (4G/5809). Blank adapter, Part No. ST11/18250. Adapter (4F/1808). Trolley, Mk 1C (4F/1714). Sleeve, AGS2111/B (28F/5722). Cone plug, AGS1140/B (28F/9439950). Adapter 4F/2459. Adapter 26DK/95401.		
<u>42251/43151</u>		
1. PREPARATION		
1.1 Pitot/static system.	Ensure all pipelines in pressurized areas are connected or blanked.	
1.2 Cockpit fittings.	Ensure installed and secure.	
1.3 Cockpit external access panels.	Ensure secure.	
1.4 Emergency ram air valve.	Ensure closed.	
1.5 Demisting control lever.	Ensure set to OFF.	
1.6 Canopy seal test point (cockpit).	Fit adapter (26DK/95437) and gauge (4F/2191).	
1.7 Canopy seal guard.	Remove.	
1.8 Hydraulic jack safety strut.		
1.9 Forward equipment hatch.	Open.	
1.10 AI23 Bullet.	Ensure fitted.	
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1. PREPARATION (contd)

1.11 Cabin/bullet exhaust nut (nosewheel bay). (i) Remove.
(ii) Fit blank, Part No. ST11/18250.

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2. PREPARATION

2.1 External d.c. power supply. (i) Connect.
(ii) Set to ON.

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3. PREPARATION

3.1 Static bleed (forward equipment compartment, Frame 4A). Blank, using sleeve (AGS2111/B) and cone plug (AGS1140/B).

3.2 Cockpit pressure test connexion (forward equipment compartment, Frame 4A). (i) Remove blank.
(ii) Fit 0 to 10 lbf/in², gauge (4G/5809) and adapters (4F/2459 and 26DK/95401) ensuring that the connecting tube is as short as possible.

3.3 Access Panel 20P (left). Remove.

3.4 Ground test connexion; Access Panel 20P (left). Connect trolley Mk 1C (4F/1714) and adapter (4F/1808).

3.5 Canopy seal inflation connexion; Access Panel 25P (left). (i) Connect inflation pump (4G/3743).
(ii) Pressurize system to approximately 100 lbf/in² as shown on gauge adjacent to inflation point.

3.6 Canopy. (i) Close, using aircraft hand-pump.
(ii) Lock, using external handle.

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3. PREPARATION (contd).

3.7 Canopy indicator lamp.

Ensure extinguished.

3.8 Canopy seal reducing valve.

Ensure operating correctly by checking that gauge (4F/2191) indicates between 7.75 and 9.5 lbf/in².

4. TESTING

4.1 Test trolley relief valve.

Set to 5 lbf/in².

4.2 Cockpit.

Pressurize at rate not exceeding 2 lbf/in² per minute, using test trolley.

4.3 Combined exhaust valve (forward equipment bay).

Check that safety valves relieve when pressure is between 4.4 and 4.5 lbf/in² as indicated on test gauge (4G/5809) on Frame 4A.

4.4 Test trolley.

Shut off.

4.5 Cockpit.

Check and record time taken for pressure to drop from 4 to 2 lbf/in².

NOTE: This should be not less than 30 secs.

4.6 Static bleed (forward equipment compartment, Frame 4A).

Remove blank.

4.7 Test trolley relief valve.

Ensure set to 5 lbf/in².

4.8 Cockpit.

Pressurize at rate not exceeding 2 lbf/in² per minute, using test trolley.

4.9 Test trolley.

Shut off.

4.10 Cockpit.

Check and record time taken for pressure to drop from 4 to 2 lbf/in².

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4. TESTING (contd)

4.11 Comparator graph
(See Fig.1).

- (i) Transfer reading obtained at Sub-item 4.5 to Y axis.
- (ii) Extend horizontally to curve of graph.
- (iii) Note point of intersection relative to X axis.
- (iv) Reading obtained at Sub-item 4.10 is to be more than that obtained at Sub-item 4.11 (iii) above.

NOTE: If reading obtained at Sub-item 4.10 is less than that obtained at Sub-item 4.11 (iii) the cabin AI bullet exhaust joint is to be treated as suspect and investigated.

5. GENERAL

5.1 Test trolley.

Disconnect.

5.2 Cabin/bullet exhaust
(nosewheel bay).

- (i) Remove blank, Part No. ST11/18250.
- (ii) Fit nut hand-tight.
- (iii) Lock with wire.

5.3 Ground test connexions:-

(a) Cockpit pressure test; Access Panel 20P (left).

Fit blank.

(b) Test connexion on Frame 4A.

- (i) Remove gauge and adapter.
- (ii) Fit blank.
- (iii) Lock with wire.

5.4 Canopy seal inflation connexion; Access Panel 25P (left).

- (i) Remove inflation pump.
- (ii) Fit Schrader cap.

5.5 Cockpit.

- (i) Ensure pressure is zero.
- (ii) Open canopy.

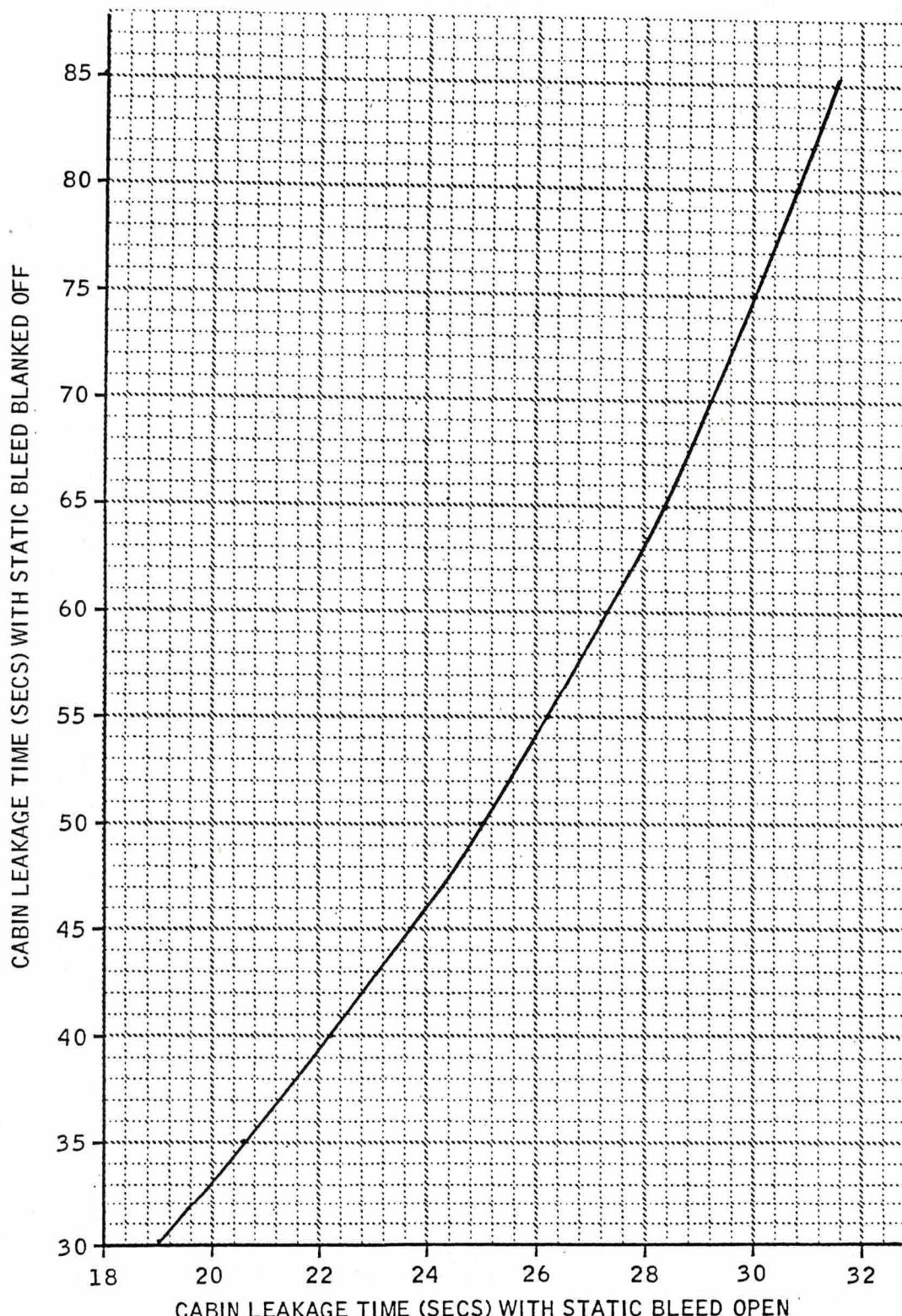
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SHEET 5 OF 6

SERVICING PROCEDURE
T55

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5A3A Section 1
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CABIN LEAKAGE

FIGURE 1

Continued Overleaf

SERVICING PROCEDURE INSPECTION STAGES DO NOT EXCLUDE ADDITIONAL INSPECTION STAGES
INCORPORATED AS NECESSARY IN MAINTENANCE CERTIFICATION DOCUMENTS

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5. GENERAL (contd)

5.6 Canopy seal test point.	(i) Remove gauge and adapter. (ii) Fit blank. (iii) Lock with wire.
5.7 Access panel 2OP (left).	Refit.
5.8 Access panels and hatches.	Ensure secure and flush fitting.

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6. GENERAL

6.1 External d.c. power supply.	(i) Set to OFF. (ii) Disconnect.
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